

Dynamic Analysis of Sandwich Beam with Viscoelastic Core Under Moving Load

Sabiiha TEKILI*, Youcef KHADRI**, Yacine KARMI***

*Université Badji Mokhtar BP12- Annaba, Laboratoire LMI, Algérie, E-mail: tekili.sabah@gmail.com

**Université Badji Mokhtar BP12- Annaba, Laboratoire LMI, Algérie, E-mail: khadri152000@yahoo.fr

***Université Badji Mokhtar BP12- Annaba, Département de Mécanique, Algérie, E-mail: karmi.yacine92@gmail.com

crossref <http://dx.doi.org/10.5755/j01.mech.26.4.23956>

1. Introduction

The knowledge of dynamic characteristics is very necessary and useful in the design and rehabilitation of structures equipped with viscoelastic materials, because due to their specific mechanical properties, viscoelastic materials contribute to the attenuation of structural vibrations. Structures by viscoelastic materials are often used in various dynamic loading conditions and in several areas during the last decades. Therefore, the viscoelastic materials development in the structures requires the implementation approaches characterizing their vibration behavior. Many research works have studied the structures vibrations under moving dynamic loads. This type of load generally causes vibrations that can extinguish the performance of the structure. These studied are focused on the impact of different dynamic parameters [1-4]. Based on the Bernoulli beam theory, a three-dimensional finite element model was proposed by Kiral et al. [5] to characterize the dynamic response of a clamped-clamped beam under dynamic load. Fuh-Gwo Y. [6] have proposed a vibration analysis model for multilayer laminated composite beams under moving load. Kahya V. [7] employed the Timoshenko theory to take into account the shear effect in each layer of composite. Sarvestan V. et al. [8] employed a quadratic equation for the moving load in order to determine the vibratory responses of the cracked Timoshenko beam under moving load. Nevertheless, the viscoelastic materials can be used as damping materials because of their specific properties. Generally, few researchers interested in linear and nonlinear viscoelastic sandwich vibrations under moving load. Irazu L. et al. [9] investigate the effect of geometric parameters on the dynamic behavior of thin sandwich structures with viscoelastic core. Billasse et al. [10] presents a numerical approach using finite element bases solution for viscoelastic sandwich vibrations. In this work, the dynamic responses of viscoelastic sandwich beams under moving load were obtained. The Hamilton principle is applied to obtain the motion equation while the finite element method is applied for the problem discretization. Different configurations of the sandwich beams and with various loss factor are considered in this study.

2. Mathematical model

The viscoelastic sandwich beam studied in this work is composed of a viscoelastic layer placed between two elastic layers constituting the sandwich skins; the beam is the length L and the width b (Fig.1). The displacement and linear strain fields of the i th face layer are given by [10]:

$$\begin{aligned} U_i(x, z, t) &= u_i(x, t) - (z - z_i)w' \\ W_i(x, z, t) &= w(x, t) \quad i = 1, 3 \\ \varepsilon_i(x, z, t) &= u_i'(x, t) - (z - z_i)w'' \end{aligned} \quad (1)$$

where: U_i and W_i are the axial displacement and the transverse displacement in the face layers for any (x) location, respectively. u_i represents the axial displacement at the centroid of the i th layer and w is the common transverse displacement. The subscripts $i = 1, 3$ refer to the upper and lower layers respectively.

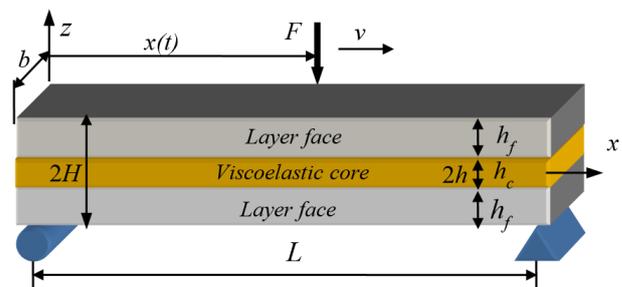


Fig. 1 Geometry of a viscoelastic sandwich beam

$z_1 = \frac{1}{2}(h_f + h_c) = -z_3$, corresponding to the middle planes ordinates of the upper and lower layers respectively. h_f and h_c are thickness of the face layers and of the viscoelastic core, respectively. The notations (g' , g'') stand for the spatial derivatives ($\partial g/\partial x$, $\partial^2 g/\partial x^2$). Taking into account the shear in the viscoelastic layer, displacement and linear strain fields of the core layer are given by:

$$\begin{aligned} U_2(x, z, t) &= u(x, t) - z\beta(x, t) \\ W_2(x, z, t) &= w(x, t) \\ \varepsilon_{n2}(x, z, t) &= u'(x, t) + z\beta' \\ \varepsilon_{s2}(x, z, t) &= \beta(x, t) + w' \end{aligned} \quad (2)$$

where: U_2 and W_2 are the axial displacement and the transverse displacement in the core layer for any (x) location, respectively. β and u corresponds to the displacement and the rotation normal of the viscoelastic layer's mid-plane, respectively. ε_{s2} is the shear strain of viscoelastic layer and ε_{n2} the normal deformation. The subscript $i=2$ refers to the core layer. The equations describing the relationship between the viscoelastic (u , β) and face layers (u_1 , u_3) are given by [10]:

$$u_1 = u + \frac{1}{2}(h_c\beta - h_f w'), \quad u_3 = u - \frac{1}{2}(h_c\beta - h_f w'). \quad (3)$$

The formulation of the motion equation of the viscoelastic sandwich beam is described by the Hamilton principle. The strain energy of the beam can be expressed as follows [11]:

$$V = \sum_{i=1}^3 \left[\frac{1}{2} \int_v (\sigma_{ij} \varepsilon_{ij}) dv \right], \quad (4)$$

where: σ_{ij} and ε_{ij} are stress and strain, respectively. The kinetic energy of the beam is:

$$T = \sum_{i=1}^3 \left[\frac{1}{2} \int_v (\rho_i \dot{u}_i \dot{u}_i + \rho_i \dot{w}_i \dot{w}_i) dv \right], \quad (5)$$

in which ρ_i is the density of the i th layer and the notation \dot{g} represents the first partial derivative of time ($\partial g / \partial t$). The virtual work of the moving load can be expressed as follows:

$$W = \sum_{i=1}^3 \left[\int_v F(x, t) w dv \right], \quad (6)$$

The moving load $F(x, t)$ is simply given by:

$$F(x, t) = F_0 \delta(x - s(t)), \quad (7)$$

where: $\delta(\cdot)$ is the Dirac-Delta function, F_0 refers to the constant force and $s(t) = vt$ is the function describing the moving of the load F_0 at time t , with v is the speed of the moving load. By using the Hamilton's principle and Eqs. (4), (5), and (6), and with the terms of axial inertia and the axial excitation force neglected, the bending motion equation can be obtained [10]:

$$\begin{aligned} & \int_0^L \{ w' \delta w' + M_\beta \delta \beta' + M_w \delta w'' + T(\delta w' + \delta \beta) \} dx = \\ & = \int_0^L F_0 \delta(x - s(t)) \delta w dx - (2\rho_f S_f + \rho_c S_c) \int_0^L \ddot{w} \delta w dx \\ & M_\beta = I_c Y * \dot{\beta}' + \frac{E_f S_f h_c}{2} (h_c \beta' - h_f w'') \\ & M_w = E_f \left(2I_f + \frac{S_f h_f^2}{2} \right) w'' - \frac{E_f S_f h_c h_f}{2} \beta' \\ & T = \frac{S_c}{2(1 + \nu_c)} Y * (\dot{\beta} + \dot{w}') \end{aligned} \quad (8)$$

where: T is shear force and M_β and M_w are bending moment. ρ_f and ρ_c are mass density of the face layers and viscoelastic core, respectively. S_f and S_c are cross-section area of the face layers and viscoelastic core, respectively. Y and ν_c are the relaxation function and the Poisson ratio of the viscoelastic material, respectively. I_f and I_c are second moment of the face layers and viscoelastic core, respectively. The notation \ddot{g} stand for temporal derivative ($\partial^2 g / \partial t^2$). No analytical solution of Eq. (8), this equation is a nonlinear differential equation; only approximated solutions will be investigated. In this paper, the approximate solution of the linear bending equation (8) is obtained by coupling the one-mode Galerkin approximation with the harmonic balance method [10]:

$$w(x, t) = W(x) e^{i\omega t}, \quad \beta(x, t) = B(x) e^{i\omega t}. \quad (9)$$

3. Finite element discretization

The finite element method will be used for the numerical discretization of the Eq. (8). An element with two nodes is used in this study, at each node there degrees of freedom (the transverse displacement w , the rotation of the normal of the central layer β and the rotation w') as shown in the Fig. 2.

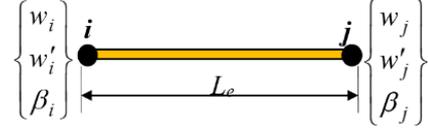


Fig. 2 Two-node sandwich beam element

The displacement field expression as a function of the nodal displacements make:

$$\{q^e\} = \{w_i \quad w'_i \quad \beta_i \quad w_j \quad w'_j \quad \beta_j\}^T, \quad (10)$$

Using classical polynomial shape functions [12], the element displacement field vector is written as:

$$\{w\} = \begin{bmatrix} N_w \\ N_\beta \end{bmatrix} \{q^e\}, \quad (11)$$

where: $[N_w]$ and $[N_\beta]$ are the interpolation functions matrices given by [12]:

$$\begin{bmatrix} [N_w] \\ [N_\beta] \end{bmatrix} = \begin{bmatrix} N_{w1} & N_{w2} & 0 & N_{w3} & N_{w4} & 0 \\ 0 & 0 & N_{\beta5} & 0 & 0 & N_{\beta6} \end{bmatrix}, \quad (12)$$

with $N_{w1} = 1 + 2 \frac{x^3}{L_e^3} - 3 \frac{x^2}{L_e^2}$, $N_{w2} = \left(\frac{x^3}{L_e^3} - 2 \frac{x^2}{L_e^2} + \frac{x}{L_e} \right) L_e$,

$N_{w3} = -2 \frac{x^3}{L_e^3} + 3 \frac{x^2}{L_e^2}$, $N_{w4} = \left(\frac{x^3}{L_e^3} - \frac{x^2}{L_e^2} \right) L_e$,

$N_{\beta5} = \left(1 - \frac{x}{L_e} \right)$, $N_{\beta6} = \left(\frac{x}{L_e} \right)$, for $x \in [0 \quad L_e]$.

Inserting Eqs. (9)- (12), into Eq. (8), one gets the following frequency–amplitude equation:

$$([K^e(\omega)] - \omega^2 [M^e]) \{q^e\} = \{F^e\}, \quad (13)$$

where: $[M^e]$ and $[K^e]$ are element mass and stiffness matrices, respectively, given by [10]:

$$\begin{aligned} [K^e] &= \frac{1}{2} \left\{ (2I_c E_c(\omega) + E_f I_f h_c^2) [k_1^e] - E_f S_f h_f h_c [k_2^e] \right\} + \\ &+ \frac{1}{2} \left\{ (E_f (4I_f + S_f h_f^2) [k_3^e] + \frac{S_c E_c(\omega)}{(1 + \nu_c)} [k_4^e]) \right\}, \end{aligned} \quad (14)$$

where: $[m^e] = \int_0^{L_e} [N_w] [N_w] dx$; $[k_1^e] = \int_0^{L_e} [N'_\beta] [N'_\beta] dx$;

$[k_2^e] = \int_0^{L_e} \left([N'_\beta] [N_w] + [N_w] [N'_\beta] \right) dx$; $[k_3^e] = \int_0^{L_e} [N_w] [N_w] dx$;

$$[k_4^e] = \int_0^{L_e} \left\{ \begin{array}{l} {}^T [N'_w] [N'_w] + {}^T [N_\beta] [N'_w] + {}^T [N'_w] [N_\beta] + \\ {}^T [N_\beta] [N_\beta] \end{array} \right\} dx.$$

The $\{F^e\}$ is element force vector:

$$\{F^e\} = F_0 \int_0^{L_e} \delta(x-vt)^T [N_w] dx. \quad (15)$$

These matrices are assembled to get the overall complex nonlinear eigenvalue problem:

$$([K(\omega)] - \omega^2 [M]) \{Q\} = 0, \quad (16)$$

where: $[K]$ and $[M]$ are the assembled mass and stiffness matrices, respectively and $\{Q\}$ is the vibration amplitude. The Eq. (16) can be solved by different methods of the eigenvalue problem as described in [13]. The amplitude equation (16) leads to an approximate value of the complex eigenfrequency that can written in the following classical form:

$$\omega^2 = \frac{K}{M} = \Omega^2 (1 + i\eta), \quad (17)$$

where: η is the structure loss factor and Ω is the structure linear frequency. The stiffness matrix K is complex with a real K^R and an imaginary K^I parts. The viscoelastic frequency and the loss factor of the viscoelastic structure are related to the real and imaginary parts of the linear stiffness by the following relationship:

$$K = K^R + iK^I. \quad (18)$$

The imaginary K^I part resulting only from the core layer (viscoelastic). For each eigenvalue, the ratio of the imaginary part to the real part is the loss factor and the square root of the real part represents the natural frequency:

$$\Omega^2 = \frac{K^R}{M}, \quad \eta = \frac{K^I}{K^R}. \quad (19)$$

This will explain the physical meaning of the real and imaginary parts of the modal stiffness constants. The introduction of an equivalent structural loss factor as a function of the amplitude makes it possible to characterize the dissipation capacity of the structure. From Eq. (16), the equation of motion for beam can be written in the form:

$$[M]\{\ddot{q}\} + [C]\{\dot{q}\} + [K]\{q\} = \{F\}, \quad (20)$$

where: $[C] = \frac{[K]^I}{\omega}$ is the equivalent damping matrix and $\{F\}$ is the external nodal load vector, obtained by assembling the formulated element load vector, defined as:

$$\{F\} = F_0 \left\{ 0 \quad \dots \quad \underbrace{N_{w1} \quad N_{w2} \quad N_{w3} \quad N_{w4}}_{\text{element under loading}} \quad \dots \quad 0 \right\}^T. \quad (21)$$

4. Results and discussions

In this study, a numerical code of finite elements is developed under MATLAB for dynamic analysis of sandwich beam with viscoelastic core under moving loads. The differential equation (20) is solved by using the implicit time integration Newmark method with parameters $\gamma = 0.5$ and $\beta = 0.25$. The mobile load is 100 N moving at a constant speed of 10 m/s. The number of finite elements used in this code is equal to 100. This code is used to calculate the frequencies, the loss factor, the dynamic responses due to the mobile load and the dynamic amplification factor. The model of viscoelastic behavior is considered with a module E_c independent of the frequency given by:

$$E_c = E_0 (1 + i\eta_v), \quad (22)$$

where: E_0 is the real modulus and η_v is the material loss factor do not depend on the frequency. This model is widely used to study viscoelastic behavior. In this paper, the Eq. (16) is solved iteratively for ω_n , the resonant frequency for mode n , and η_n , the corresponding modal loss factor.

4.1. Validation

The real natural frequencies and loss factors of sandwich of the simply supported sandwich beam with viscoelastic core with aluminum face layers are calculated from Eq. (19) and compared with the corresponding results obtained by Billasse [10]. The mechanical properties of the viscoelastic sandwich beam with Aluminum face layers are given in Table 1.

Table 1

Mechanical properties of the the sandwich beam.

	Elastic face	Viscoelastic core
Young's module, Pa	$E_f = 6.9E10$	$E_0 = 1749E3$
Poisson's ratio	$\nu_f = 0.3$	$\nu_c = 0.3$
Density, kg/m ³	$\rho_f = 2766$	$\rho_c = 968.1$
Thickness, m	$h_f = 1.524$	$h_c = 0.127$
Length, Width, m	$L = 177.8, \quad b = 12.7$	

The results are obtained for different values of viscoelastic core loss factor $\eta_v = 0.1, 0.6$ and 1.0 . The real natural frequencies and loss factors corresponding to the first three modes sandwich beam are given in Table 2. The results show that as the damping factor of the viscoelastic core increases, the damped pulsation also increases involving the improvement of the structural damping level.

Table 2

Real natural frequencies Ω and loss factor η of the sandwich beam for different values of η_v .

η_v	Mode	Present formulation		Billasse [10]	
		Ω , Hz	η/η_v	Ω , Hz	η/η_v
0.1	1	148.41	0.3501	148.51	0.3502
	2	488.27	0.1953	488.48	0.1958
	3	1034.50	0.1067	1034.75	0.1071
0.6	1	150.60	0.33283	150.71	0.3328
	2	489.54	0.19388	489.76	0.1943
	3	1035.20	0.10654	1035.44	0.1069
1.0	1	154.30	0.3053	154.42	0.3052
	2	491.83	0.1914	492.07	0.1918
	3	1036.40	0.1062	1036.69	0.1065

The obtained results are very acceptable, given their coincidences with those obtained by Billasse [10] that shows the effectiveness of the present approach.

4.2. Vibrations analysis

The effects of geometric parameter (h/H) and viscoelastic loss factor on natural frequencies and structure loss factors are studied. First, we examine the effect of the sandwich beam configuration parameters such as viscoelastic loss factor and the ratio of thickness h/H on the damping properties of the structure. The configuration and geometric properties of the sandwich beam considered are given in Fig. 1 and Table 3, respectively.

Table 3

Mechanical properties of the viscoelastic sandwich beam.

	Layer face	Viscoelastic core
Young's modulus(Pa)	$E_1=14.7E10$ $E_2=9.0E9$ $G=5.0E9$	$E_c=7.037E5(2(1+ \nu_c))$
Poisson's ratio	$\nu_1=0.3$	$\nu_c=00.49$
Density (kg/m^3)	$\rho_1=1580$	$\rho_c=970$
Thickness (m)	$h_f = H-h$ $h_2 = 2h$ $H = 0.012$ $h = 0.0012$	

4.2.1. Free vibrations

The variations of the real natural frequencies and loss factors corresponding to the first three modes for different configuration parameters, viscoelastic loss factor and thickness ratio of the simply supported sandwich beam are reported in Fig. 3.

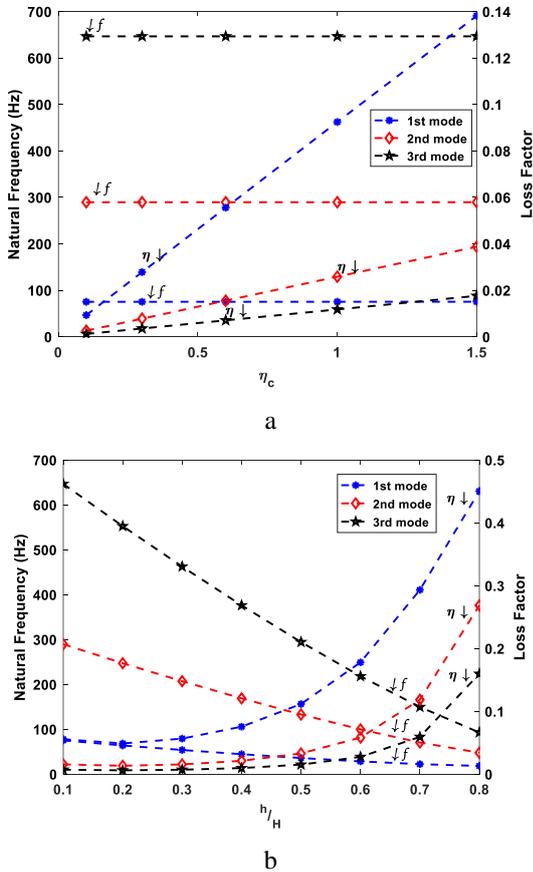


Fig. 3 Real nature frequencies and loss factors of the sandwich beam according to the parameters: a) η_v ; b) h/H

Proportional relationship between the viscoelastic loss factor and the structure loss factor shown in Fig. 3, a, nevertheless the eigen frequencies remain constant. The results are quite logical given the importance of the imaginary part of Young's modulus that reflects the ability of viscoelastic materials to dampen vibrations. Especially when this imaginary part is considerable, the results of Fig. 3, b, illustrate that the frequencies are inversely proportional to the thickness ratios h/H . This means that the frequencies decrease when the thickness ratio increases.

4.2.2. Forced vibrations

The time-forced responses of the sandwich beam are illustrated for different values of η_v . Fig. 4 show the responses dynamic obtained for different values of $\eta_v = 0.1, 0.6, 1.0$ and 1.5 with $h/H=0.1$ at velocity of moving load, $v = 10$ m/s. It is observed that the effect of the viscoelastic loss factor is particularly significant for free oscillations whose amplitudes of oscillations are very small for the large values of η_v .

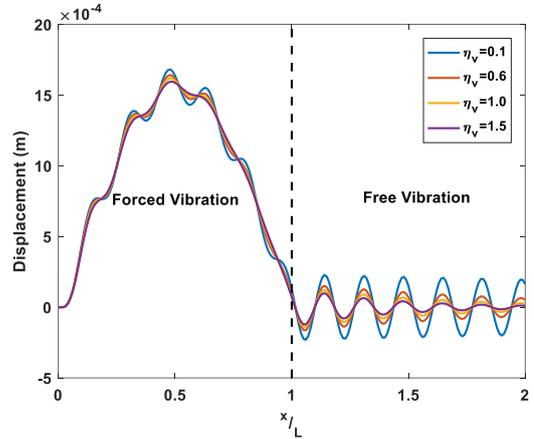


Fig. 4 Dynamic responses of the sandwich beam under moving load for different values of η_v

The dynamic amplification factor (DAF) responses for different values of η_v are reported in Fig. 5. This factor is defined as the ratio between the maximum value of the dynamic displacement under moving load and the static displacement under concentrated force. The Fig. 5 show that the critical speeds vary from one configuration to another, for which $\eta_v=0.6, V_c = 75, 25, 45$ and 25 (m/s) respectively. The impact of the viscoelastic damping on the DAF is very considerable considering the decrease in the amplitudes of the DAF of the studied cases for with $h/H = 0.1$, after having increased the viscoelastic loss factor.

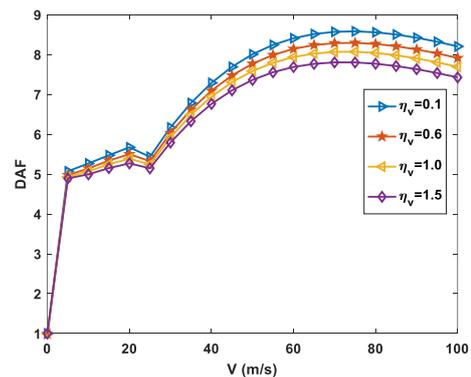


Fig. 5 DAF of the sandwich beam for different values of η_v

The deflection responses of the sandwich beam with the effect of the thickness ratio are illustrated in Fig. 6 for different values of the viscoelastic loss factor η_v at $v = 10$ m/s. The results obtained from Figs. 6, a – c reveal that the amplitudes of the responses increase with the increase in the h_c whose amplitudes with $\eta_v = 0.6$ reach $w=1.6e-3$, $3.5e-3$, $15e-3$ and $46.4e-3$ (m) for respectively $h/H=0.1$, 0.3 , 0.6 and 0.8 . However, the damping capacity becomes more significant for these values which have been noticed for the ratios $h/H = 0.6$ and 0.8 whose free oscillations have disappeared more rapidly compared with those obtained with $h/H = 0.1$ and 0.3 . The viscoelastic damping can be improved by increasing the thickness of the viscoelastic layer while keeping the thickness of the face layers.

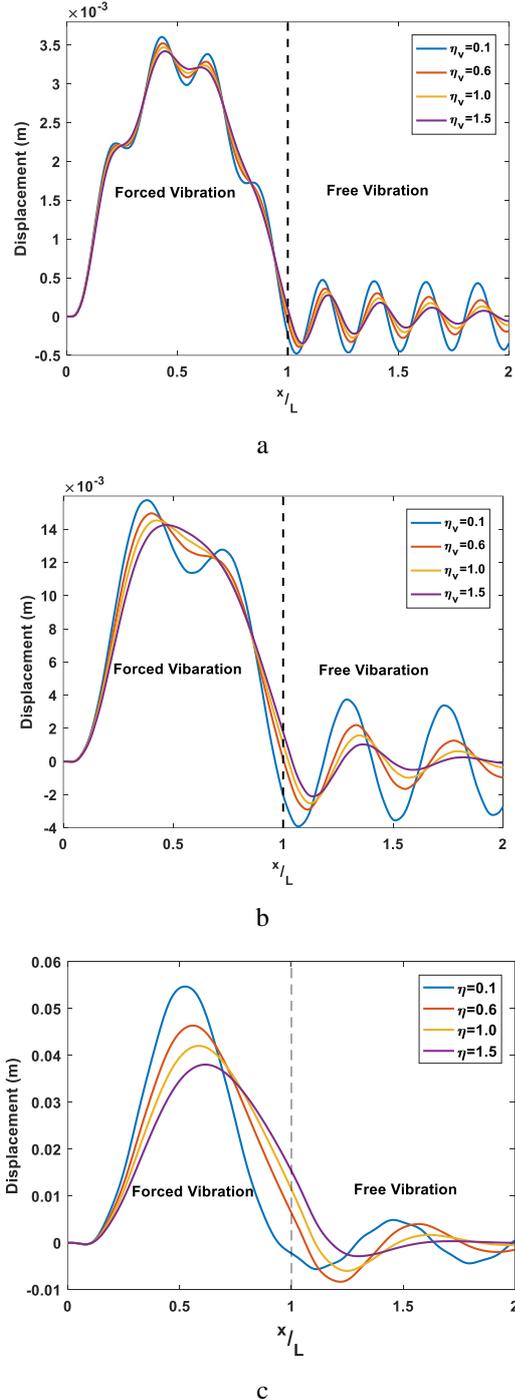


Fig. 6 Forced dynamic responses of the sandwich beam for different values of η_v : a) $h/H=0.3$; b) $h/H=0.6$; c) $h/H=0.8$

The responses obtained from the DAF for different values of η_v and h/H are presented in Fig. 7. The effect of the damping is also clear from what is illustrated in Figs. 7, a – c especially in the hypercritical region where the effect of the viscoelastic damping becomes more significant.

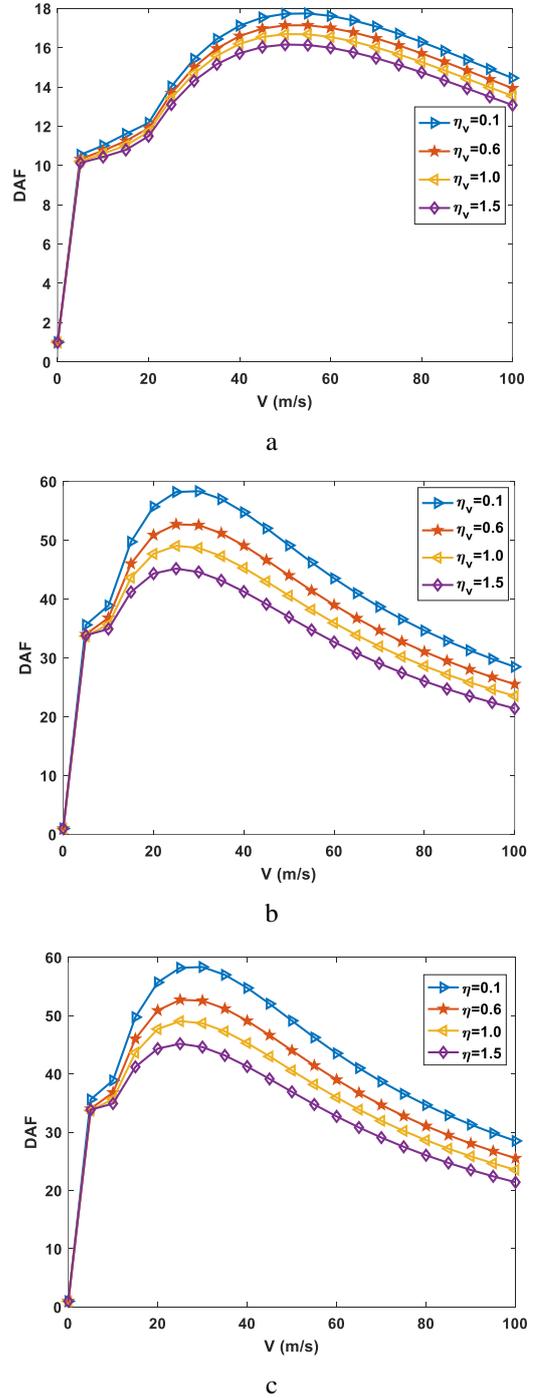


Fig. 7 DAF of the sandwich beam for different values of η_v : a) $h/H=0.3$; b) $h/H=0.6$; c) $h/H=0.8$

5. Conclusions

The vibration behaviors of sandwich beams with viscoelastic core under moving force are investigated numerically. From the work done, several conclusions can be drawn: The significant effect of the loss factor of frequency-independent viscoelastic modulus on the dynamic behavior involving the improvement of the overall damping

of the structure which consequently allows to reduce the amplitudes of vibrations when the factor becomes considerable. The maximum amplitudes of the vibratory responses of the sandwich beams are proportional to the thickness ratio H/h between the viscoelastic layer and the face layer, the damping of which becomes very significant with a very thin thickness of the viscoelastic layer involving the decrease of the amplitudes of displacement. Thus, the damping of the structure becomes more significant as the thickness of the viscoelastic core layer becomes thinner. This amounts to the shear force applied to the viscoelastic layer for this type of configuration, which consequently makes it possible to further improve the damping provided essentially by the viscoelastic layer.

Acknowledgements

The authors gratefully acknowledge the Algerian General Direction of Research for providing the facilities and the financial funding of this project (A11-N01-UN23-01-2019-0002).

References

1. **Hilal, M. A.; Zibdeh, H. S.** 2000. Vibration analysis of beams with general boundary conditions traversed by a moving force, *J. Sound Vib.* 229(2): 377–388. <https://doi.org/10.1006/jsvi.1999.2491>.
2. **Giunta, F.; Muscolino, G.; Sofi, A.; Koff, I. E.** 2017. Dynamic analysis of Bernoulli-Euler beams with interval uncertainties under moving loads, *Procedia Eng.* 199: 2591–2596. <https://doi.org/10.1016/j.proeng.2017.09.353>.
3. **Chen, Y.; Fu, Y.; Zhong, J.** et al. 2017. Nonlinear dynamic responses of fiber-metal laminated beam subjected to moving harmonic loads resting on tensionless elastic foundation, *Compos. Part B Eng.* 131: 253–259. <https://doi.org/10.1016/j.compositesb.2017.07.051>.
4. **Tekili, S.; Khadri, Y.; Merzoug, B.** 2015. Finite element analysis of free vibration of beams with composite coats, *Mechanika* 21(4): 290-295. <https://doi.org/10.5755/j01.mech.21.4.9849>.
5. **Kiral, Z.; Kiral, B. G.** 2008. Dynamic analysis of a symmetric laminated composite beam subjected to a moving load with constant velocity, *J. Reinf. Plast. Compos* 27(1): 19–32. <https://doi.org/10.1177/0731684407079492>.
6. **Fuh-Gwo, Y.; Miller, R. E.** 1989. A new finite element for laminated composite beams, *Comput. Struct.* 31(5):737–745. [https://doi.org/10.1016/0045-7949\(89\)90207-1](https://doi.org/10.1016/0045-7949(89)90207-1).
7. **Kahya, V.** 2012. Finite element dynamic analysis of laminated composite beams under moving loads, *Structural Engineering & Mechanics* 42(4):729-745. <https://doi.org/10.12989/sem.2012.42.5.729>.
8. **Sarvestan, V.; Mirdamadi, H. R.; Ghayour, M.** 2017. Vibration analysis of cracked Timoshenko beam under moving load with constant velocity and acceleration by spectral finite element method, *Int. J. Mech. Sci.* 122: 318–330. <https://doi.org/10.1016/j.ijmecsci.2017.01.035>.
9. **Irazu, L.; Elejabarrieta, M. J.** 2017. The effect of the viscoelastic film and metallic skin on the dynamic properties of thin sandwich structures, *Comput. Struct.* 176: 407–419. <https://doi.org/10.1016/j.compstruct.2017.05.038>.
10. **Bilasse, M.; Daya, E. M.; Azrar, L.** 2010. Linear and nonlinear vibrations analysis of viscoelastic sandwich beams, *J. Sound Vib.* 329 (23): 4950–4969. <https://doi.org/10.1016/j.jsv.2010.06.012>.
11. **Meirovitch, L.** 1997. Principles and techniques of vibrations, 1st ed. Prentice Hall.
12. **Clough, R. W.; Penzien, J.** 1993. Dynamics of structures, McGraw-Hill, New York, USA.
13. **Daya, E. M.; Azrar, L.; Potier-Ferry, M.** 2004. An amplitude equation for the non-linear vibration of viscoelastically damped sandwich beams, *Journal of Sound and Vibration* 271:789–813. [https://doi.org/10.1016/S0022-460X\(03\)00754-5](https://doi.org/10.1016/S0022-460X(03)00754-5).

S. Tekili, Y. Khadri, Y. Karmi

DYNAMIC ANALYSIS OF SANDWICH BEAM WITH VISCOELASTIC CORE UNDER MOVING LOAD

S u m m a r y

In this article, numerical approach is proposed for dynamic behavior of symmetrical sandwich beams with viscoelastic core under moving load using the Hamilton's principle formulation and the finite element method solution. The dynamic responses are obtained for different configurations using the modal superposition method and the implicit Newmark integration scheme. The analysis shows that the viscoelastic damping has a significant effect on the vibration behavior involving the improvement of the damping of the structure. The parametric study of the effect of the configuration parameters shows that the sandwich structure has more dissipative capacities of vibratory energy by adopting adequate configurations to the structure.

Keywords: vibration, sandwich, viscoélastic material, damping, moving load.

Received: September 26, 2019

Accepted: August 24, 2020



This article is an Open Access article distributed under the terms and conditions of the Creative Commons Attribution 4.0 (CC BY 4.0) License (<http://creativecommons.org/licenses/by/4.0/>).